

XFoil  
V 6.9

-2.0  
-1.5  
 $C_p$   
-1.0  
-0.5  
0.0  
0.5  
1.0

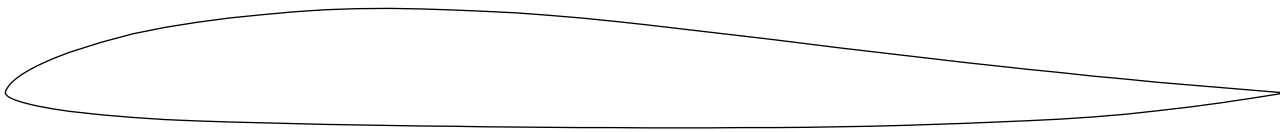
BW 05 02 09

$\alpha = 2.5000^\circ$

$C_L = 0.3140$

$C_M = 0.0175$

$C_{Dp} = -0.00037$



XFOIL  
V 6.9

BW 05 02 09 (SM00 01)

$\alpha = 2.5000^\circ$

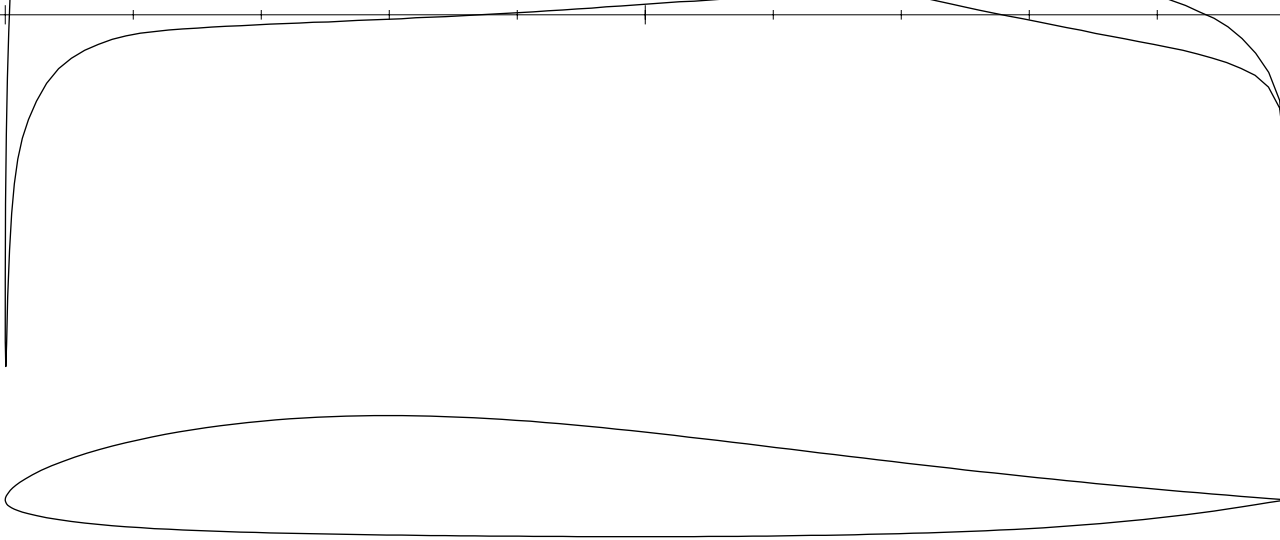
$C_L = 0.3031$

$C_M = 0.0189$

$C_{Dp} = -0.00018$

$C_p$

-2.0  
-1.5  
-1.0  
-0.5  
0.0  
0.5  
1.0



XFOIL  
V 6.9

BW 05 02 09 (SM00 01)

Re =  $0.300 \times 10^6$

$\alpha$  =  $2.5000^\circ$

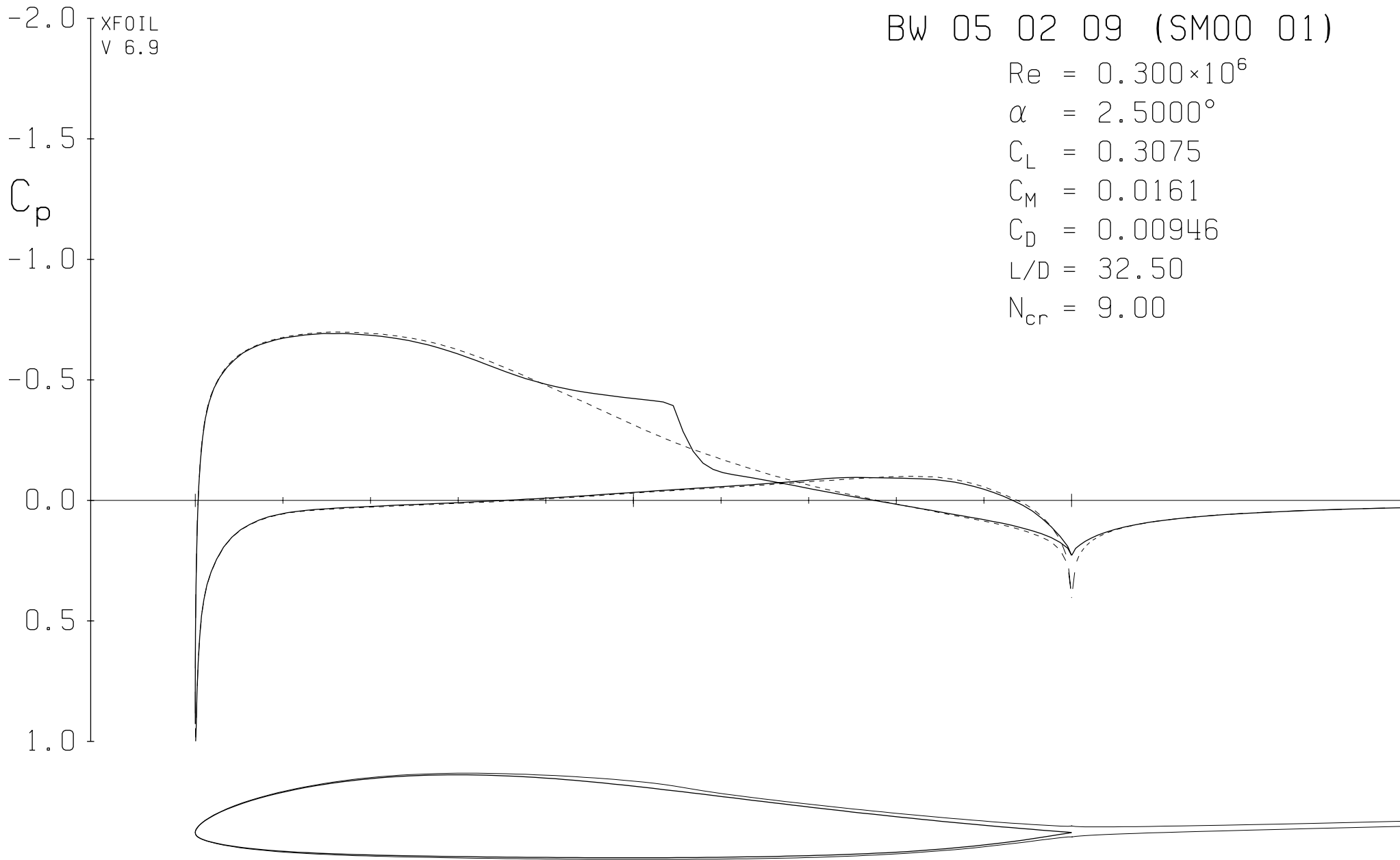
$C_L$  = 0.3075

$C_M$  = 0.0161

$C_D$  = 0.00946

L/D = 32.50

$N_{cr}$  = 9.00



XFOIL  
V 6.9

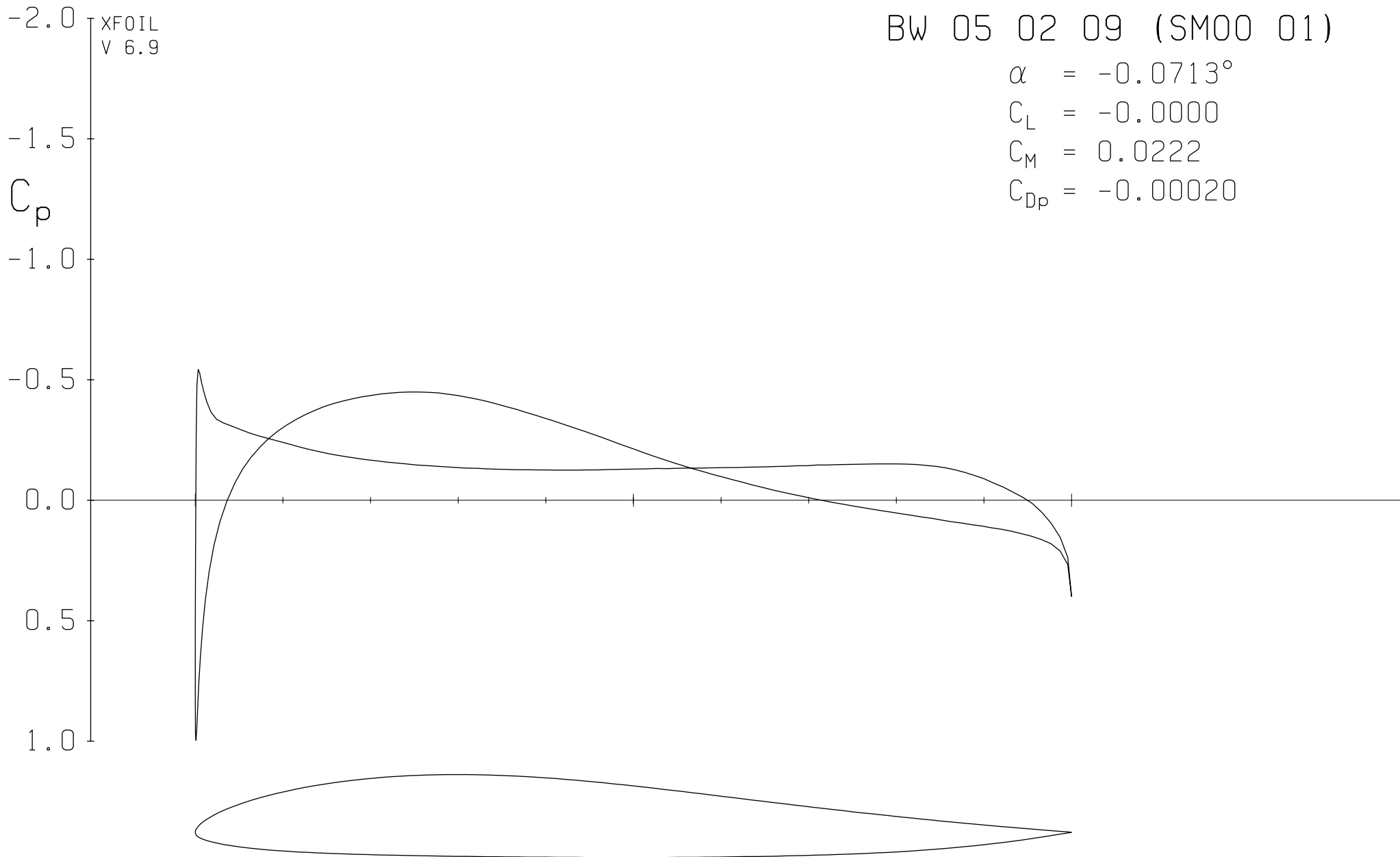
BW 05 02 09 (SM00 01)

$\alpha = -0.0713^\circ$

$C_L = -0.0000$

$C_M = 0.0222$

$C_{Dp} = -0.00020$



XFOIL  
V 6.9

BW 05 02 09 (SM00 01)

Re =  $0.300 \times 10^6$   
 $\alpha = -0.3270^\circ$   
 $C_L = -0.0000$   
 $C_M = 0.0143$   
 $C_D = 0.00844$   
L/D = -0.00  
 $N_{cr} = 9.00$

